

METHOD FOR REPAIRING GAS TURBINE  
COMPRESSOR ROTOR BLADES

A method facilitates repairing a gas turbine engine compressor blade airfoil. The method comprises removing titanium alloy material from along leading and trailing edges of the airfoil, and along a radially outer tip of the airfoil to form respective leading edge, trailing edge, and tip cut-backs which each define cut-back depths, and depositing titanium weld material onto the leading edge, trailing edge, and tip cut-backs. The method also comprises removing at least some of the titanium weld bead material to obtain pre-desired finished dimensions for the leading and trailing edges, and radially outer tip.